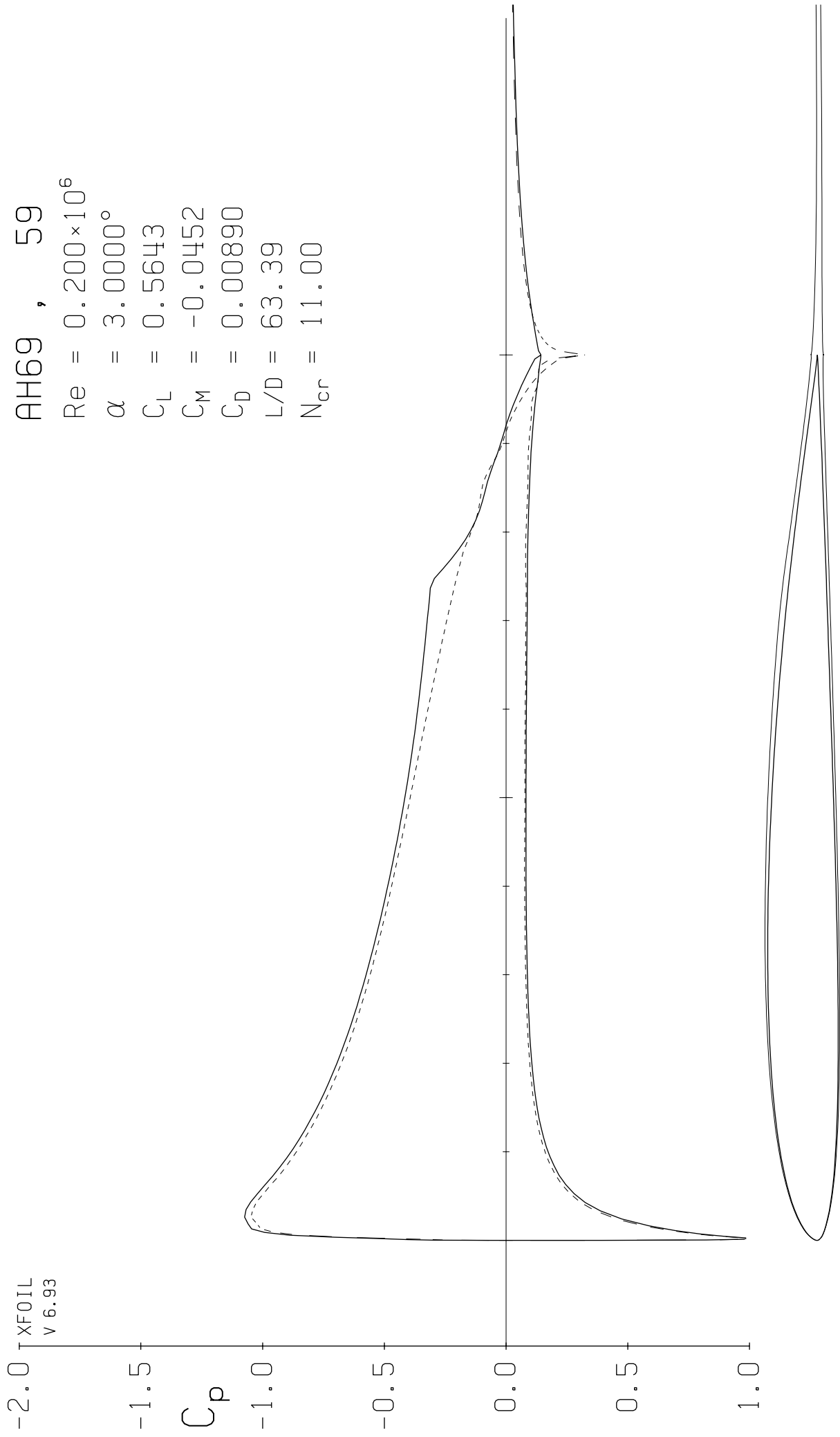


-2.0 XFOIL
V 6.93



AH69 , 59

$Re = 0.200 \times 10^6$

$\alpha = 3.0000^\circ$

$C_L = 0.5643$

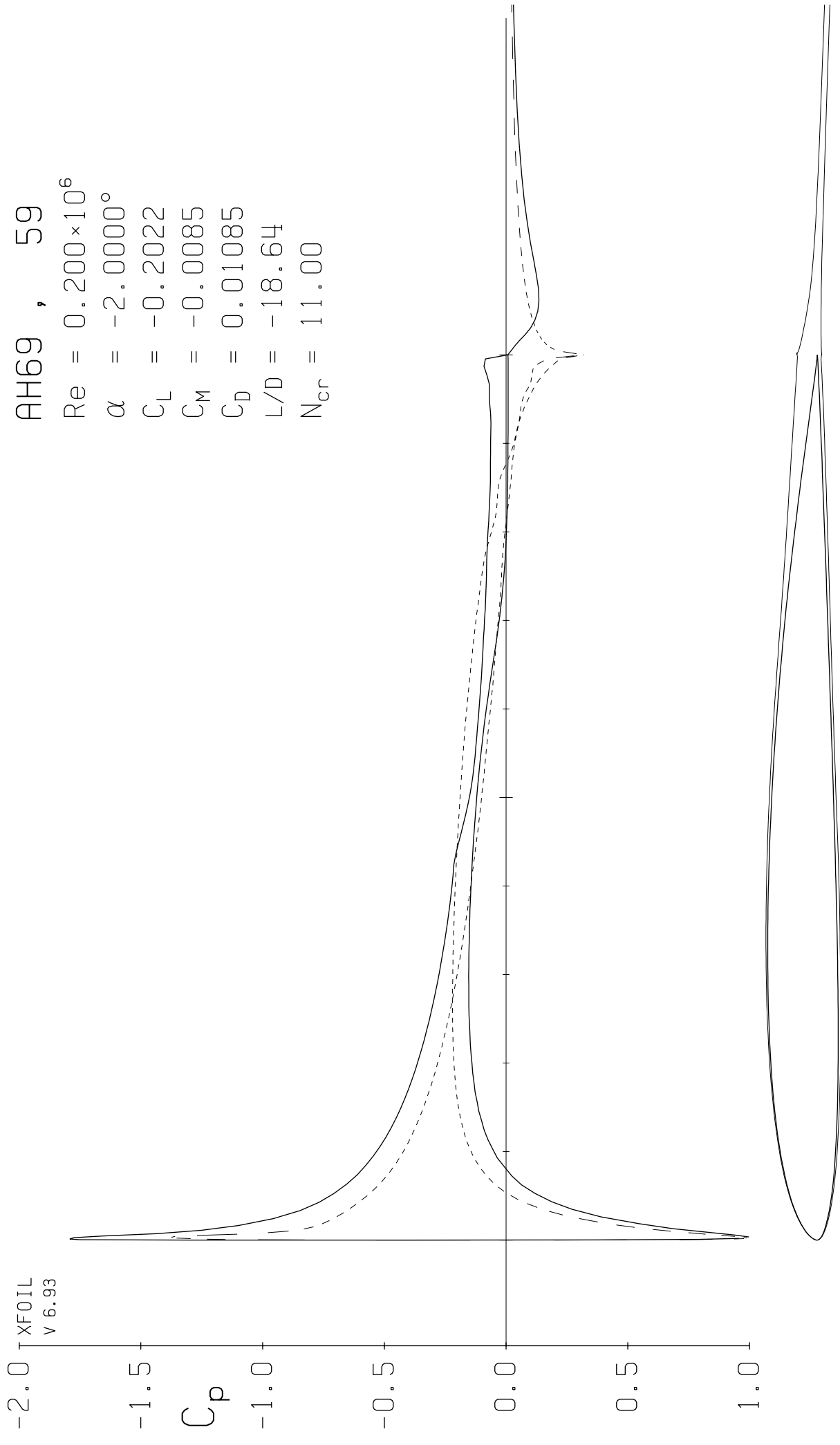
$C_M = -0.0452$

$C_D = 0.00890$

$L/D = 63.39$

$N_{cr} = 11.00$

-2.0 XFOIL
V 6.93



AH69 , 59

Re = 0.200×10^6

α = -2.0000°

C_L = -0.2022

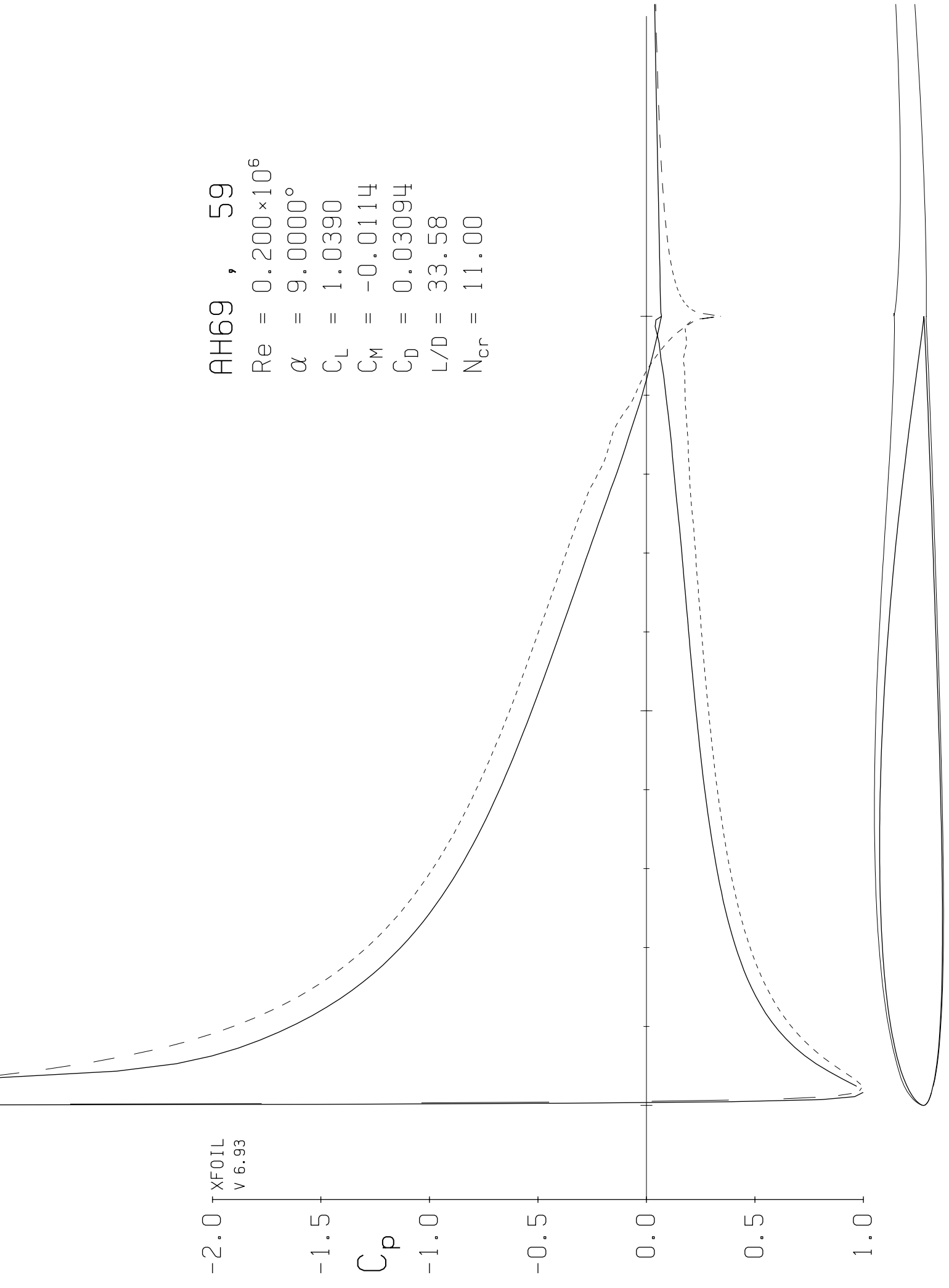
C_M = -0.0085

C_D = 0.01085

L/D = -18.64

N_{cr} = 11.00

AH69 , 59
Re = 0.200×10^6
 $\alpha = 9.0000^\circ$
 $C_L = 1.0390$
 $C_M = -0.0114$
 $C_D = 0.03094$
L/D = 33.58
 $N_{cr} = 11.00$



-2.0 XFOIL
V 6.93

AH69 , 59
Ma = 0.000
Re = 0.200 × 10⁶
N_{cr} = 11.000

